

ABSTRACT

Noise reduction conduit for non rotary components of aircraft engines, subjected to a characteristic range of temperatures of a gas turbine engine, constituted by an  
5 annular structure composed of an aerodynamic wet wall (10a, 10b), perforated and resistant mechanically and thermally; of a dry wall (12a, 12b), not resistant and of light weight; and of some intermediate elements to which both walls are mechanically attached and that define a  
10 jump or difference of temperature between the wet and dry walls; between which wet (10a, 10b) and dry walls (12a, 12b) there are partitions that define cavities (16a, 16b) isolated with regard to each other.